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A. B. BLOSE
Technical Information Officer

DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS ABOUT AIRCRAFT WITH APPLICATION TO EXTERNAL STORES

VOLUME IV. - MACH NUMBER 1.025 FLOW-FIELD SURVEY DATA FOR THE 4-PERCENT THICK WING-BODY COMBINATION

> By Stanley C. Perkins, Jr., Stephen S. Stahara and Michael J. Hemsch

> > NEAR TR 138 July 1977

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NIELSEN ENGINEERING & RESEARCH, INC. Mountain View, California

for

AIR FORCE OFFICE OF SCIENTIFIC RESEARCH Bolling Air Force Base Washington, D. C.

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EXTERNAL STORES

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20. ABSTRACT (Construe on reverse side if necessary and identify by block number)

A test program was conducted to obtain measurements of flow velocities and static pressures in the vicinity of wing-body-store model (representative of a fighter-type aircraft) as well as surface pressures, forces, and moments on the model. Flow velocities and static pressures were also measured near the tunnel walls to provide outer flow field information. This report presents the data obtained during the test program conducted in the 4T and 16T Wind Tunnels at Arnold Engineering Development Center. The Flow-field data were obtained at Mac numbers 0.925, 0.975, and 1.025 and constitute the major part of the data. (cont

Volume I is a summary report which gives detailed information on the test program and presents uncertainties associated with the various types of data taken in the 4T Wind Tunnel. The volume also presents tunnel-empty and Mach-number surveys, as well as tabulated force and moment and pressure data for the Mach number range 0.80 to 1.15 and angles of attack -2°, -5°, 0°, 2°, and 5°. Volumes II, III, and IV present the tabulated flowfield data for the 4-percent thick wing model at Mach numbers 0.925, 0.975 and 1.025, respectively. Volume V presents the tabulated flow-field data for the 6-percent thick wing model, and Volume VI presents data obtained for the 4-percent thick wing model in the 16T Wind Tunnel.

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7	

### NOMENCLATURE

This section provides a list of symbols which identify various aerodynamic parameters, axis designations, subscripts, and tabulated data nomenclature.

alverate-model angle of attack, positive none up

. . as seen by the pilot (does down is turnel), dec

### Symbols

烈性过度.

t fix on probe	betafublic installation emissing incol	40
AAL	local upwash angle, deg; $tan^{-1}(\frac{WL/VM}{UL/VM})$	area
M	Mach number negrous construction to the control of	
entreliga sun ovair edk	radius of the body, in.	214
Re/ft	free-stream Reynolds number per foot, f	E_1
SWL paisms	local sidewash angle, deg; tan (VL/VM)	on as
UL,VL,WL	local velocity components, positive alone positive X, Y, and Z directions, reft/sec	ng the spectively,
ending decar.	free-stream velocity, ft/sec	. versa
x,Y,Z	body-fixed Cartesian coordinates with o coincident with the aircraft model nos angles of attack, see figure 4(a)	rigin e at all
	tunnel-fixed Cartesian coordinates with coincident with the aircraft model nos angle of attack, see figure 4(b)	origin e at zero
a section o	angle of attack of the model, deg; angle body axis and tunnel axis, as defined	e between in figure 4
90	azimuthal angle in the Y-Z plane, deg from the positive Y axis as shown in	; measured figure 4
	Subscripts	
	for because a conditions and proper	394
	Gegy tan (VICUL)	
e ins	Flow Field Survey Data Tabulation	2753

deg; tan (WL/UL)

AAL

local upwash angle referenced to body-axis

coordinates, calculated from probe measurements,

### NOMENCLATURE (Continued)

AATL väidmebi	
aubacripts,	
ALFA	aircraft-model angle of attack, positive nose up as seen by the pilot (nose down in tunnel), deg
CPL	local pressure coefficient calculated from probe measurements, (PL -P)/Q
DATE	calendar time at which data were recorded
M	wind tunnel free-stream Mach number
ML.	local Mach number calculated from probe measurements
P	free-stream static pressure, psfa
PART	sequential indexing number for referencing data; a constant throughout each survey
PLowledges.	local static pressure calculated from probe measurements, psfa
POINT  Alektro di Lis de ecou	sequential indexing number for referencing data obtained during one part; indexes each time a new set of data inputs is obtained.
PT	wind tunnel free-stream total pressure, psfa
PTD LEGIC MALE	local total pressure measured by probe, psfa
Ω	wind tunnel free-stream dynamic pressure, psf
REX10-6 of this	wind tunnel free-stream unit Reynolds number, millions per foot
RUN 1910 toob	identifier for specific user test type
SURVEY	identifier for specific user grid-survey combina- tion
SWL .	local sidewash angle referenced to body-axis coordinates, calculated from probe measurements, deg; tan (VL/UL)
SWTL sixo-yb	sidewash angle referenced to tunnel-axis coordinates, calculated from probe measurements, deg; tan (VT/UT)
· · · · · · · · · · · · · · · · · · ·	dogs tan (wirth)

### NOMENCLATURE (Concluded)

I	TEST	alpha-numeric notation for referencing a specific test unit
	TT	wind tunnel free-stream total temperature, OF
	UL,VL,WL	velocity components in the body-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec
Ī	UT,VT,WT	velocity components in the tunnel-axis X, Y, and Z directions, respectively, calculated from probe measurements, ft/sec
п	VM	wind tunnel free-stream velocity, ft/sec
1	VML	local velocity calculated from probe measurements, ft/sec
Ш	WING	wing designation used for a specific part number
п	<b>x</b>	location of the probe in the body-axis X direction
п	XT	location of the probe in the tunnel-axis X direction
) II	<b>Y</b> .	location of the probe in the body-axis Y direction
11	YT	location of the probe in the tunnel-axis Y direction
П	2	location of the probe in the body-axis Z direction
1	<b>2T</b>	location of the probe in the tunnel-axis Z direction

#### DATA REPORT FOR A TEST PROGRAM TO STUDY TRANSONIC FLOW FIELDS ABOUT AIRCRAFT WITH APPLICATION TO EXTERNAL STORES

VOLUME IV. - MACH NUMBER 1.025
FLOW-FIELD SURVEY DATA FOR THE
4-PERCENT THICK WING-BODY
COMBINATION

## 1. INTRODUCTION

This volume of the data report presents the flow-field survey data at  $M_{\infty}=1.025$  for the 4-percent thick wing-body model. The data were obtained in the 4T Wind Tunnel at Arnold Engineering and Development Center. These tests, performed at a nominal Reynolds number per foot of  $3.0\times10^6$ , are outlined in Tables I through VI of this volume. The tabulated data are at the end of this volume beginning on page number 1.

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stationerscored oritions. Columns electe and a

#### 2. DESCRIPTION OF TESTS

The details of the test hardware, coordinate systems, and data uncertainties associated with these tests, as well as an overview of the purpose and scope of the test program, are provided in Volume I of this data report. Figure 1 of this volume shows the entire grid layout used for the inner flow-field measurements. Some layouts use only a portion of this pattern, depending on Mach number and angle of attack. The general grid layout used for the outer flow-field measurements is shown in figure 2. Two different grid layouts were used depending on the angle of attack. The grid layout is designed to give outer flow-field data at a constant radial distance from the tunnel centerline for various values of 0, the azimuthal angle. A sketch of the wing-body combination is shown in figure 3.

The flow-field survey tests at Mach number 1.025 are summarized in Table I. Columns one through three indicate the angle of attack, the table number, and the page numbers, respectively, for each set of data.

Tables II through VI contain the test condition grids for

each angle of attack. Columns one and two indicate the page number and part number, respectively, of the tabulated data for each flow-field run. Columns four, five, and six indicate the initial and final positions and the incremental change in the axial coordinate, the body-axis X direction, of the probe static-pressure orifices. Columns eight and nine indicate the lateral and vertical coordinates, the body-axis Y and Z directions, respectively, of the probe longitudinal centerline. These coordinates indicate the various inner flow-field surveys. The outer flow-field surveys are found on the second page of each table and at the end of each table. Columns four, five, and six of these surveys indicate the initial and final positions and incremental change, respectively, of the axial coordinate, the tunnel-axis X direction, of the probe static-pressure orifices. Columns eight and nine indicate the lateral and vertical coordinates, the tunnel-axis Y and Z directions, respectively, of the probe longitudinal centerline. Column ten indicates the azimuthal angle of each outer flow-field run, as defined in figure 4 of this volume.

# 3. DESCRIPTION OF DATA

The flow-field survey data for the 4-percent thick wing-body model at Mach number 1.025 are presented in tabular form on pages 1 through 220 at the end of this volume. The heading on each page contains the test number, the part number, the Reynolds number per foot, the angle of attack of the model, the type of wing attached to the model (4-percent thick wing for this volume), and the Y and Z (or YT and ZT) coordinates at which the X (or XT) traverse is carried out. Also included are the run and survey numbers and the date on which data were recorded.

Below the heading information are the data obtained during each test. Column one indicates the sequential indexing number for referencing data obtained during one part (POINT). Column two

indicates the location of the probe static-pressure orifices in the body-axis X direction (X) for the inner flow-field surveys or in the tunnel-axis X direction (XT) for the outer flow-field surveys. Columns three through seven indicate wind tunnel free-stream quantities. These are Mach number (M), velocity (VM, ft/sec), total pressure (PT, psfa), dynamic pressure (Q, psf) and total temperature (TT, oF). Columns eight through sixteen indicate local quantities which were either measured by the probe or calculated from probe measurements. Columns eight through eleven contain the local Mach number (ML), the ratio of local to free-stream velocity (VML/VM), the ratio of local to free-stream total pressure (PTL/PT), and the local pressure coefficient (CPL). For the inner flow-field surveys, columns twelve through sixteen contain the ratio of local velocity components in the body-axis X, Y, and Z directions, respectively, to the free-stream velocity (UL/VM, VL/VM, and WL/VM, respectively) and the local upwash and sidewash angles (AAL and SWL, respectively) referenced to body-axis coordinates. For the outer flow-field surveys, columns twelve through sixteen contain these same local quantities as determined in the tunnel-axis coordinate The positive sense of the velocity components is along the positive X, Y, and Z directions. A positive local upwash angle indicates downward flow away from the wing-body combination, the positive Z or ZT direction, see figure 5. A positive local sidewash angle indicates flow along the positive Y or axis, see figure 5.

indicates the location of the probe static-provide omifices in the body-axis X direction (X) for the inner flow-field surveys of in the twendi-axis X direction (XT) for the outer flow-field enryeys. Columns three through seven I direct wind threat free-street quantities. These are much mider (M),

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### TABLE I. - FLOW-FIELD SURVEY TESTS

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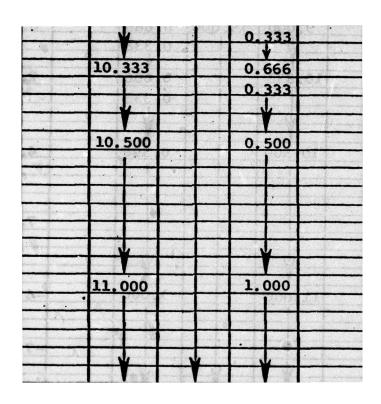
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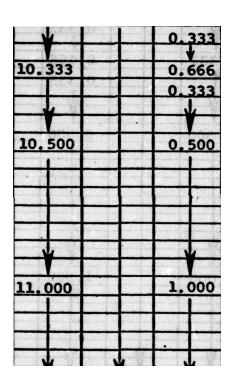
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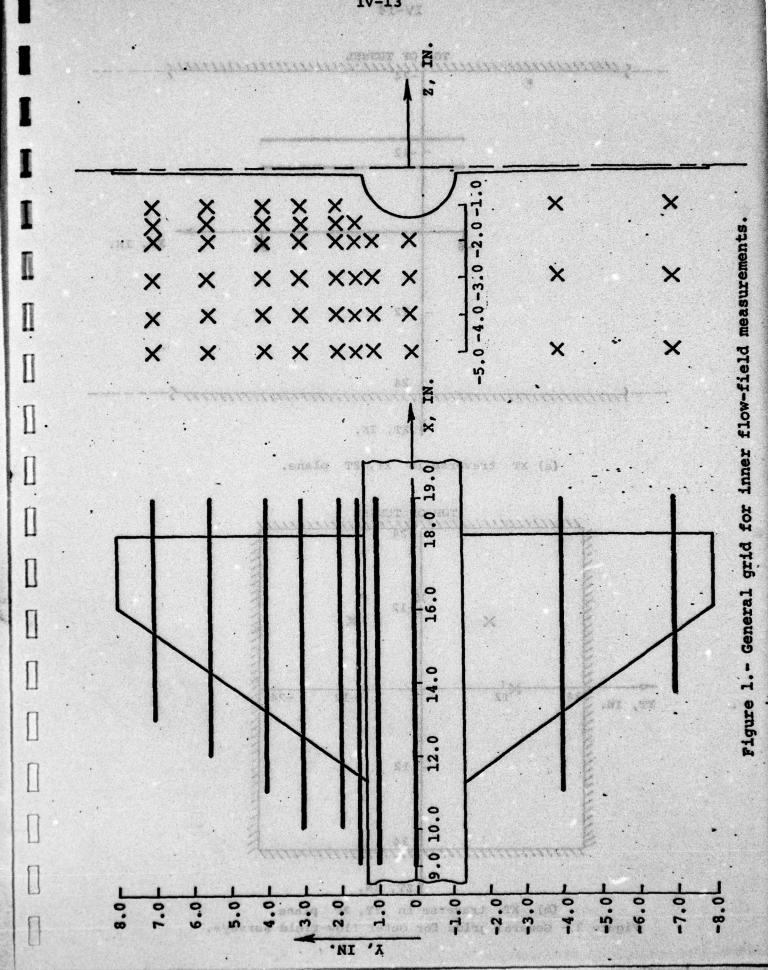
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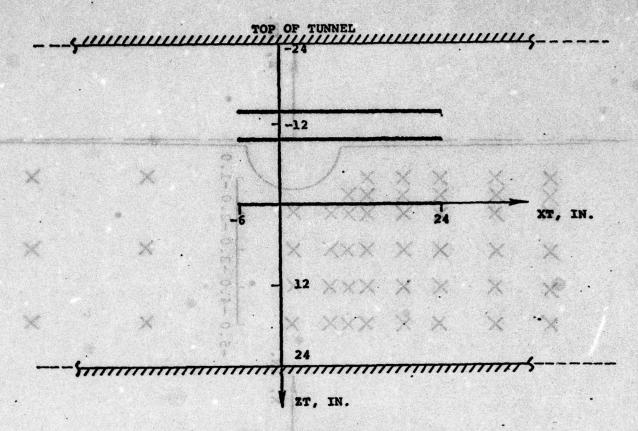
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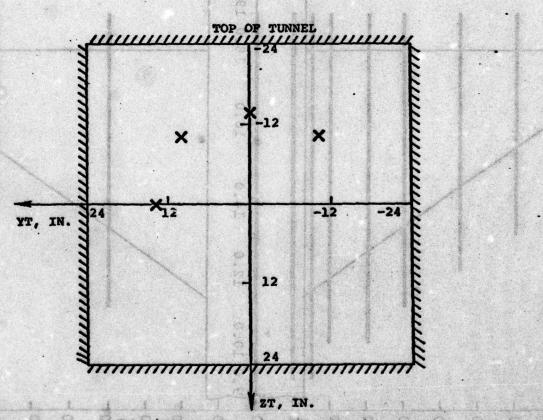




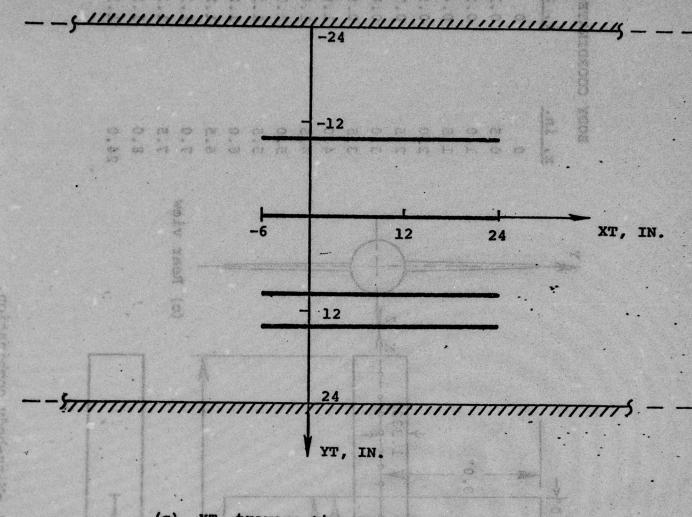




(a) XT traverse in XT, ZT plane.



(b) XT traverse in YT, ZT plane. Figure 2.- General grids for outer flow-field surveys.



(c) XT traverse in XT, YT plane.

Figure 2.- Concluded.

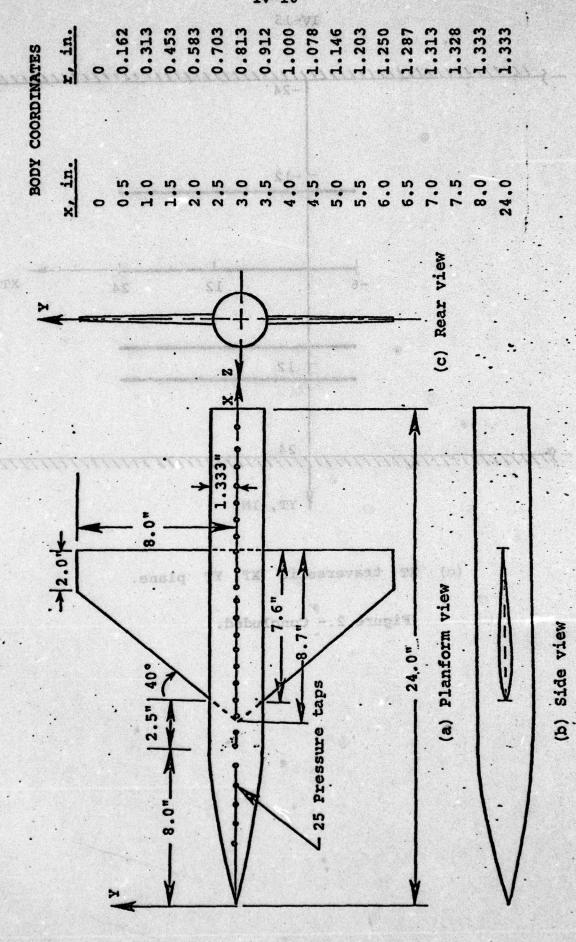
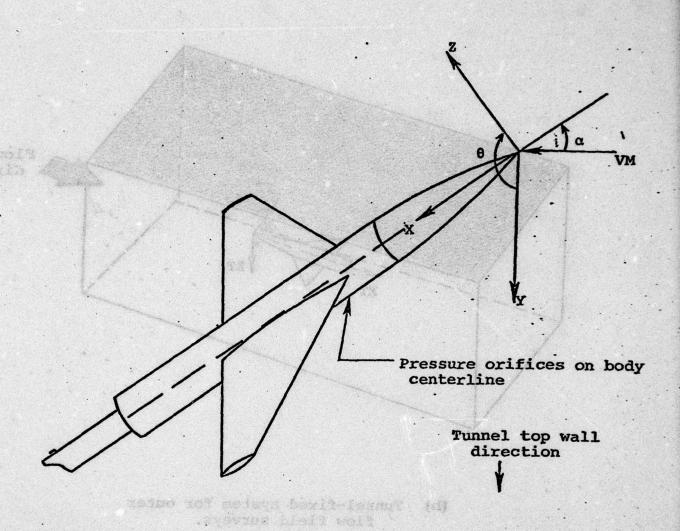


Figure 3.-Wing-body combination.

数主用製造

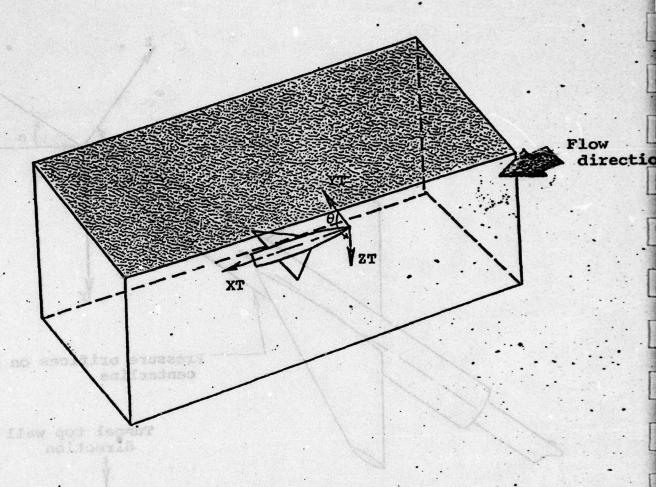


(a) Body-fixed system for inner flow field surveys.

Figure 41- Conditions

Figure 4.-Coordinate systems.

マエージル



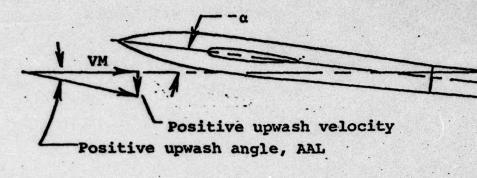
(b) Tunnel-fixed system for outer flow field surveys.

Figure 4.- Concluded.

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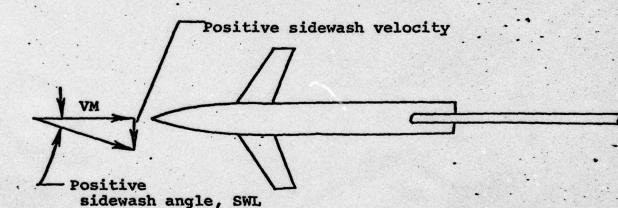
. Pigoco 4, - Coordinate systems. .

TOP WALL OF TUNNEL



BOTTOM WALL OF TUNNEL

(a) Side view of tunnel.



WALL OF TUNNEL

(b) Plan view of tunnel from top.

Figure 5.-Pictorial sign convention for upwash and sidewash angles.

### ### ### ### ### ### ### ### ### ##	1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1,			1,022 1058,41 1399,1 1,023 1059,61 1399,1 1,025 1061,46 1399,7		PUR SUPPEY			la .	DATE	VEDC	PROPUL	610 wind	TURNEL
1,000   1,00	1,000   1,00			9,000 1,022 1058,41 1399,8 9,333 1,023 1059,61 1399,6	000	012-0								3
10.00	10.00	100   100	1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1,	9,666 1,025 1061,46 1349.7	126.0 00.0	1:103	];	4-	]_	100.	\$ 8.		6.33	-
Color   Colo	1,000	Color   Colo	1,000	2000 1001 6701	527.5 79.9	1-116	===			056 0	100		0.45	0.1
100   101	Color	Color	Column   C	10.000 1.026 1062.31 1399.6	528.6	1.092	22		5	052	.003			0.21
100   100	1.07   100.1   1191	1.077   1001.11   1371.6   572.0   10.01   0.011   0	Color	10.166 1.026 1061.55 1398.9	126 0 60 0	1.036	9	2 =	1	000	8	000	0.03	0
1,000   1,00		Color   Colo		11.000 1.026 1061.79 1398.6	528.0 80.0	1.007	2.2	==	- 5	985	88	2.00	00.0	
1025   1016_6   1199_6   228_6   1000   0.00	1021   1011.65   1391.65	Color   Colo	100   100	11.666 1.627 1062.96 1399.5	526.9	0.987		-	•	196	88	6000	-0.51	
0.995 0.996 1.001 0.069 0.997 0.003 -0.009 -0.598 0.003 -0.009 -0.598 0.003 -0.009 -0.598 0.003 -0.009 -0.598 0.003 -0.009 -0.009 0.995 0.995 0.003 -0.009 -0.009 0.995 0.995 0.003 -0.009 -0.009 0.995 0.995 0.995 0.003 -0.009 -0.009 0.995 0.995 0.995 0.003 -0.009 -0.009 0.995 0.995 0.995 0.003 -0.009 -0.009 0.995 0.995 0.995 0.003 -0.009 0.995 0.995 0.995 0.003 -0.003 -0.009 0.995 0	0.985 0.967 1.001 0.069 0.966 0.003 0.005 0.005 0.966 0.003 0.006 0.966 0.003 0.006 0.966 0.003 0.006 0.966 0.003 0.006 0.967 0.003 0.006 0.967 0.003 0.006 0.967 0.003 0.006 0.967 0.003 0.006 0.967 0.003 0.006 0.967 0.003 0.005 0.005 0.003 0.005 0.005 0.003 0.005 0.005 0.003 0.005 0.005 0.003 0.005	0.985 0.986 1.001 0.069 0.986 0.003 -0.009 -0.51 0.985 0.985 0.986 1.001 0.067 0.987 0.003 -0.009 -0.51 0.987 0.985 0.986 0.986 0.986 0.987 0.987 0.987 0.987 0.987 0.987 0.987 0.987 0.987 0.003 -0.006 -0.98 0.987 0.987 0.987 0.987 0.997 0.003 -0.006 -0.98 0.987 0.987 0.997 0.003 -0.007 -0.007 0.987 0.997 0.003 -0.007 0.987 0.997 0.003 -0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.987 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.997 0.997 0.007 0.99	0.915 0.905 1.001 0.005 0.905 0.003 0.005 0.905	12.333 1.026 1061.65 1398.9	528.0 80.0	0.940		22				0.010	-0.57	
0.967 0.967 1.001 0.067 0.967 0.003 -0.009 -0.56 0.966 0.967 1.001 0.067 0.967 0.003 -0.006 -0.30 0.968 0.967 1.001 0.065 0.969 0.003 -0.006 -0.30 0.998 0.974 1.001 0.053 0.974 0.003 -0.007 0.998 0.974 1.001 0.053 0.976 0.003 -0.007 1.001 0.976 1.001 0.053 0.976 0.004 0.002 0.32 1.001 0.976 1.001 0.014 0.976 0.004 0.007 1.001 0.996 1.001 0.014 0.976 0.004 0.007 1.002 0.997 1.001 0.001 0.976 0.005 0.007 1.003 0.997 1.001 0.001 0.976 0.005 0.007 1.004 1.006 1.001 0.007 0.997 0.005 0.007 1.005 1.001 0.007 0.997 0.005 0.997 1.005 1.001 0.007 0.997 0.005 0.007 1.005 1.005 1.001 0.007 0.007 0.005 0.007 1.006 1.001 0.007 0.007 0.007 0.007 1.007 1.008 1.000 0.007 0.007 0.007 1.008 1.000 0.007 0.007 0.007 0.007 1.008 1.009 0.007 0.007 0.007 0.007 1.009 0.007 0.007 0.007 0.007 0.007 1.009 0.007 0.007 0.007 0.007 0.007 0.007 1.009 0.007 0.007 0.007 0.007 0.007 0.007	0.967 0.967 1.001 0.067 0.967 0.003 -0.009 -0.51 0.005 0.967 0.005 -0.009 -0.005 0.967 0.005 0.967 0.005 -0.009 -0.005 0.967 0.005 0.967 0.005 -0.005 0.967 0.005 0.967 0.005 -0.005 0.967 0.005 0.967 0.005 0.967 0.005 0.975 0.005 0.005 -0.005 0.975 0.005 0.975 0.005 0.005 0.005 0.975 0.00	0.987 0.967 1.001 0.067 0.967 0.003 -0.008 -0.51 0.001 0.005 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 -0.008 0.003 0	0.987 0.967 1.001 0.067 0.967 0.003 -0.009 -0.561 0.986 0.986 0.987 1.001 0.065 0.987 0.003 -0.008 -0.008 0.986 0.986 0.987 1.001 0.065 0.987 0.003 -0.008 -0.008 0.987 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.995 0.003 -0.007 -0.008 0.995 0.99	12.666 1.027 1062.69 1398.4	528.3 50.1	0.985	66	===	89	996	.003	0.000	-0.55	
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1.054 1.006 1.001 -0.011 1.006 0.005 0.012 0.70 1.050 1.018 1.001 -0.034 1.017 0.005 0.017 0.93 1.051 1.005 1.001 -0.034 1.017 0.005 0.017 0.93 1.051 1.012 1.001 -0.037 1.019 0.005 0.012 0.93 1.085 1.085 1.000 -0.083 1.032 0.005 0.025 1.38 1.085 1.085 1.000 -0.083 1.084 0.005 0.027 1.38 1.087 1.083 1.000 -0.083 1.031 0.006 0.021 1.38	1.034 1.006 1.001 -0.011 1.006 0.003 0.012 0.70 1.023 0.097 1.001 -0.014 1.017 0.005 0.017 0.93 1.034 1.005 1.001 -0.004 1.019 0.005 0.017 0.93 1.041 1.001 -0.003 1.019 0.005 0.017 0.95 1.085 1.083 1.000 -0.083 1.084 0.005 0.025 1.46 1.085 1.083 1.000 -0.083 1.005 0.002 1.146 1.086 0.981 1.000 0.014 0.981 0.006 0.017 0.11	1.054 1.006 1.001 -0.011 1.000 0.005 0.012 0.70 1.050 1.016 1.001 -0.011 1.000 0.005 0.013 0.91 1.051 1.005 1.001 -0.014 1.017 0.005 0.017 0.91 1.051 1.012 1.001 -0.017 1.019 0.005 0.012 0.91 1.067 1.012 1.001 -0.063 1.019 0.005 0.021 1.15 1.068 1.068 1.000 -0.013 1.051 0.006 0.021 1.16 1.069 1.000 -0.013 1.005 0.027 1.16 1.000 0.013 1.000 0.013 1.014 0.005 0.013 1.16	1.054 1.006 1.001 -0.011 1.000 0.005 0.012 0.70 1.050 1.016 1.001 -0.011 1.000 0.005 0.013 0.91 1.051 1.005 1.001 -0.014 1.010 0.005 0.012 0.91 1.051 1.012 1.001 -0.017 1.019 0.005 0.012 0.91 1.051 1.012 1.001 -0.051 1.019 0.005 0.012 0.91 1.067 1.012 1.000 -0.013 1.014 0.005 0.021 1.16 1.067 1.001 1.000 -0.013 1.014 0.005 0.021 1.16 1.067 1.001 1.000 -0.014 0.005 0.013 1.16	16.933 1.026 1062.22 1399.8	528.7 79.9	1.0	000	5	22	::	5000	0.0	-	0.7
1.023 0.997 1.001 0.007 0.997 0.005 0.009 0.53 1.014 1.005 1.001 -0.008 1.004 0.005 0.002 1.051 1.013 1.000 -0.013 1.005 0.002 1.085 1.005 -0.013 1.005 0.025 1.005 1.095 1.001 1.000 -0.013 1.005 0.002 1.096 0.013 1.000 0.003 1.014 0.005 0.002 1.005 1.000 0.003 1.000 0.003 1.014 0.005 0.002 1.005	1.023 0.997 1.001 0.007 0.997 0.005 0.009 0.53 1.034 1.003 1.001 -0.009 1.004 0.005 0.003 0.93 1.095 1.001 -0.003 1.005 0.004 0.005 0.005 1.095 1.005 0.005 1.005 0.005 0.005 1.15 1.095 1.005 0.005 1.005 0.005 1.40 1.095 1.005 0.005 1.005 0.005 1.40 1.095 1.005 0.005 1.40 1.007 0.005 0.0	1.023 0.997 1.001 0.007 0.997 0.005 0.009 0.53 1.034 1.005 1.001 -0.009 1.004 0.005 0.002 1.041 1.001 -0.037 1.032 0.005 0.023 1.042 1.005 0.003 1.004 0.005 0.025 1.05 1.093 1.003 1.000 -0.013 1.005 0.003 1.05 1.000 0.003 1.000 0.003 1.005 0.003 1.04 1.000 0.003 1.000 0.003 1.005 0.003 1.005	1.023 0.997 1.001 0.007 0.997 0.005 0.009 0.53 1.034 1.005 1.001 0.003 1.004 0.005 0.003 0.013 1.045 1.001 0.003 1.004 0.005 0.003 1.05 1.045 1.000 0.013 1.005 0.025 1.05 1.045 1.000 0.013 1.005 0.025 1.05 1.045 1.000 0.013 1.005 0.025 1.05 1.045 0.013 1.000 0.013 0.005 0.012 0.11	17.000 1.027 1062.02 1400.0	529.0 80.1	6	90	50	=	900	0.00			0.2
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1,000   1,004   1,00		22.200				79.9	990	1.054	1.001	-0.107		-0.00	-0.001	0.00	
1,200   1,024   10,2	1,000   1,00	13.500				79.7	1.067	1.034	1.000	0.025		0.007	-0.002	20.0	
11,000 1.024 1001.22 1399.1 577.0 79.7 1.095 1.005 0.099 0.9	13,000 1.024 1.019	13.500				79.8	0.975	0.959	1.001	****		0.012	-0.00	9.4	
15.00	1,000   1,027   1002   20   1994   2374   1757   0.975   0.975   1001   0.092   0.955   0.015   0.015   0.955   0.015   0.015   0.955   0.015   0.015   0.955   0.01	li				7.00	0.971	0.957	1.001	0.0		0.015	0.00	-0.52	
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16,000 1,027 1052,22 1397,1 527.8 79.6 0.897 0.897 1.001 0.049 0.896 0.003 0.009 0.0	15.000 1.027 102.23 137.1 357.8 1.039 0.576 1.001 0.049 0.049 0.059 1.001 1.004 0.002 -0.003 -0.003 1.004 0.000 0.15 0.003 1.004 0.000 0.15 0.003 1.004 0.000 0.15 0.003 1.004 0.000 0.15 0.003 1.004 0.005 1.004	15.500				79.7	0.971	0.950	1.001	0.092		0.010	-0.014		
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15-000   1-023   1005-129   1396-5   1371-7   171-9   0.976   1-001   0.082   0.956   1-001   0.082   0.956   197-0   0.023   0.956   1700   0.023   0.959	13-000 1-023 104-13 104-14 134-1 232-1 37-2 0-37-1 0-100 0-001 0-0	$1 \cdot 1 \cdot 1$	0000			- 50.5	526.2 526.1 526.7	2222	10000		10001	0.000	1.036	-0.00 -0.00	LEADER PARTY STATES AND ADDRESS OF THE PARTY STATES AND ADDRES	40000	
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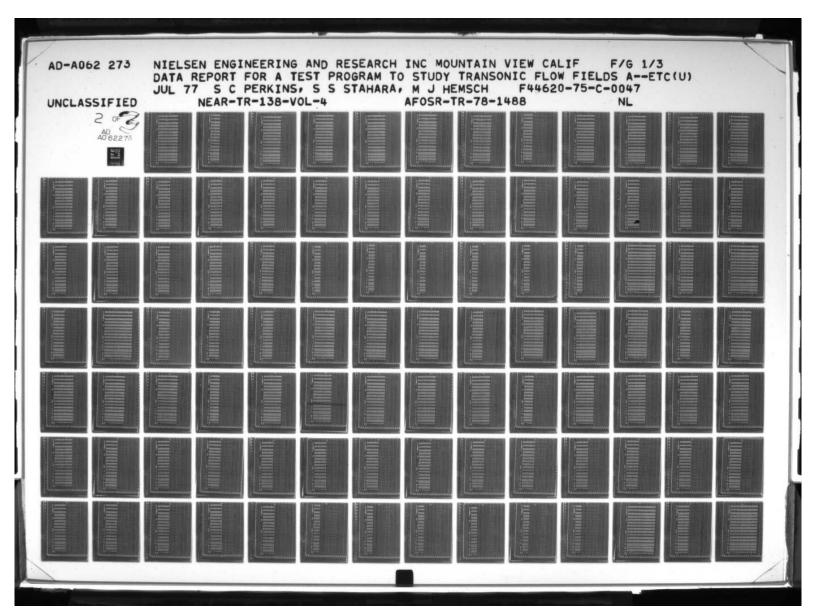
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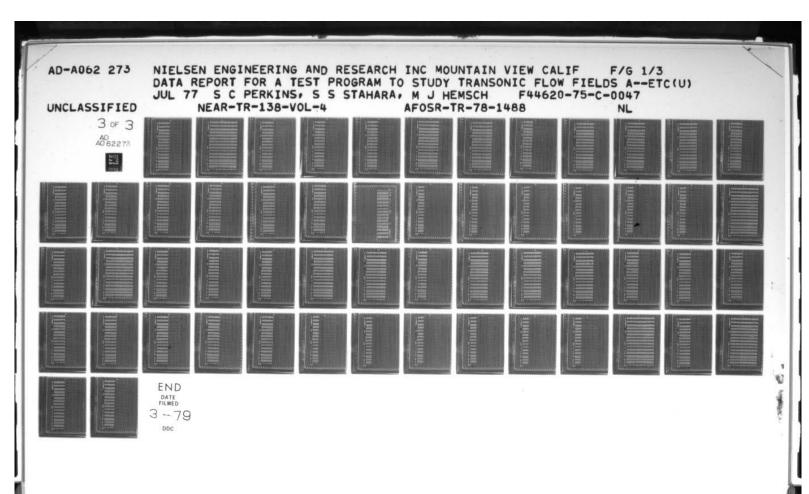
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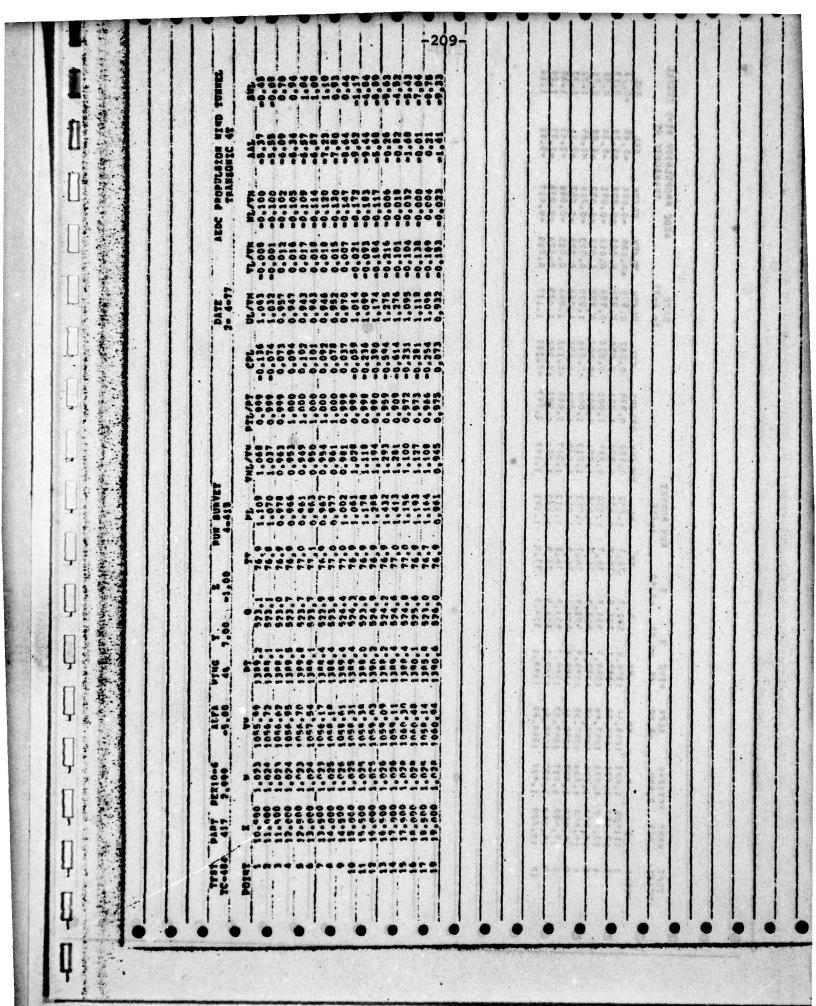
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